



SUBJECT:

Required maintenance for the RH Door Handle Installation (P/N 120-200014).

APPLICABILITY:

Aircraft with the subject modification embodied in accordance with TCCA STC. No. SH12-10 or any relevant foreign approvals.

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	NAME AND SIGNATURE	DATE	COMPANY / DEPARTMENT
PREPARED BY:	D. Kerr <i>D. Kerr</i>	2 May 2011	ECL ENGINEERING
PREPARED BY:			ECL ENGINEERING
CHECKED BY:	C. Timmins <i>C. Timmins</i>	2 nd May 2011	ECL ENGINEERING
CHECKED BY:	M. Merritt <i>M. Merritt</i>	2011-05-02	ECL QUALITY ASSURANCE
APP'D / ACCEPTED (Civil A/W Authority)	<i>C. Timmins</i> (As per ICA Compliance Check Sheet)	13 Feb 2012	TCCA
RELEASED BY:	R. Manson <i>R. Manson</i>	13 Feb 2012	ECL ENGINEERING



RECORD OF REVISIONS

Rev.	Pages at this Revision	Description, Reason, Changed Pages	Prepared (name and date)	Checked (name and date)	App'd/Acc'd (Civil A/W Authority) (name and date)	Released (name and date)
0	1 through 12	Original Issue	See page 1.	See page 1.	See page 1.	See page 1.

NOTE: Revisions to this document will be distributed to operators of this equipment by the STC holder.

NOTE: Revised portions of affected pages are identified by a vertical black line in the margin adjacent to the change.

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1. GENERAL

- A. The subject RH Door Handle Installation is installed on the inside of the forward right-hand door providing the crew member a convenient and easily accessible means of closing the RH Door.

The RH Door Handle is manufactured from high performance thermoplastic (PEI). Two door reinforcements are secured to the door channel. Two spacers are attached to the outer door reinforcement which can be adjusted to achieve the correct fit for the door handle once the door side panel is attached.

The RH Door Handle Installation consists of the following main components:

Fixed Provisions:

- inner door reinforcement
- outer door reinforcement
- spacer

Detachable Provisions:

- door handle

- B. The Instructions for Continued Airworthiness are applicable to aircraft with the subject modification embodied.

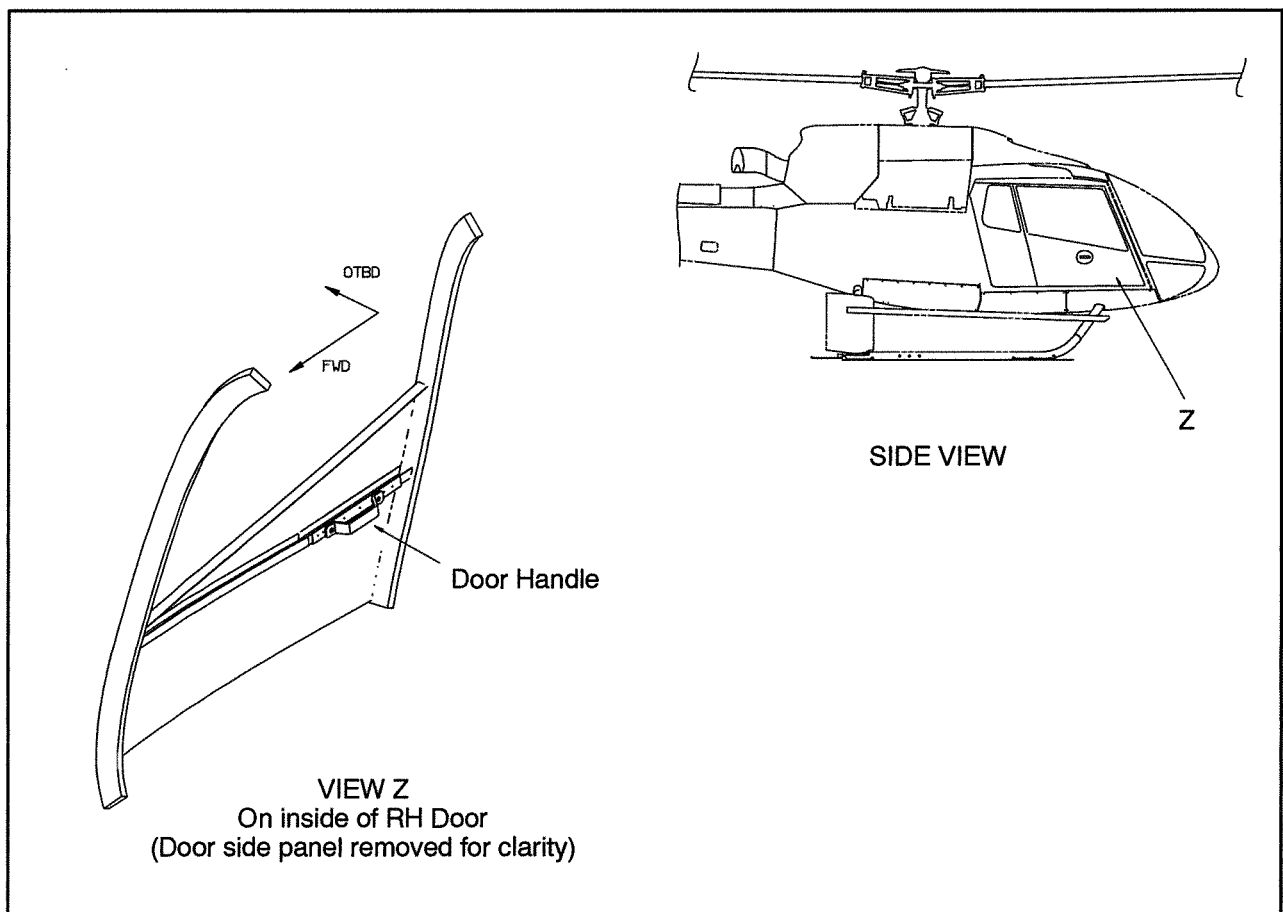


Figure 1 General Layout

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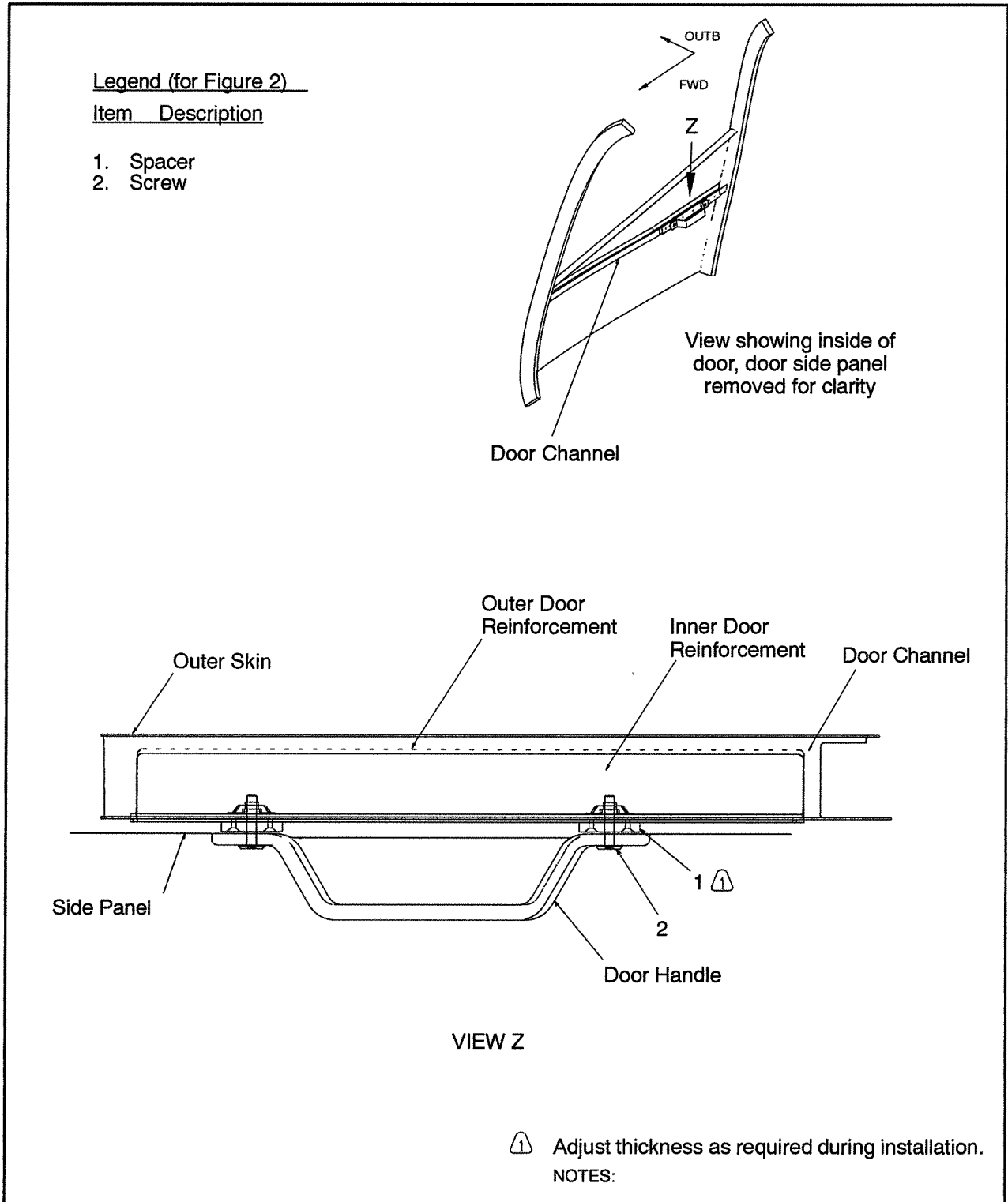


Figure 2 Door Handle Details (Sheet 1 of 2)

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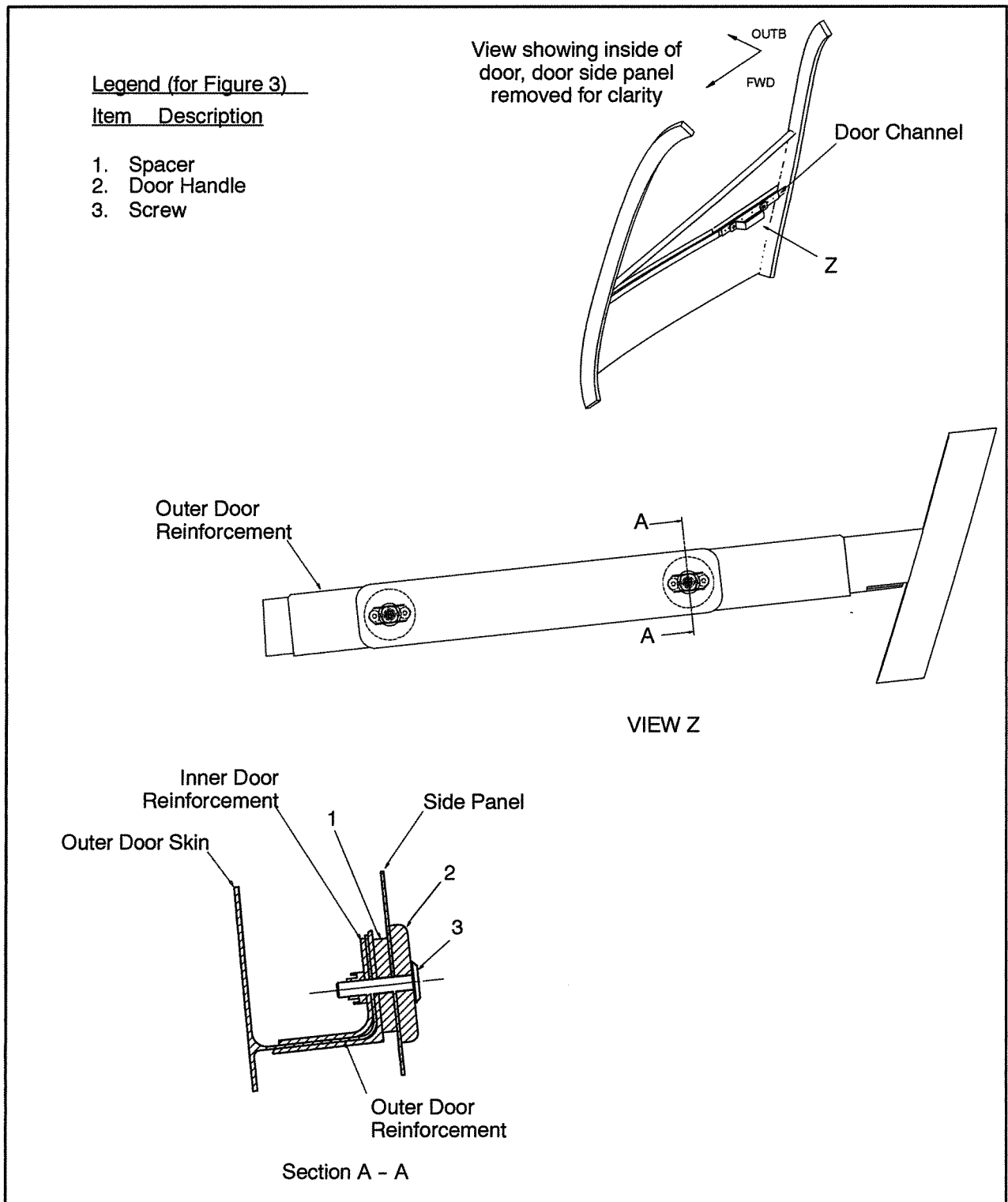


Figure 3 Door Handle Details (Sheet 2 of 2)

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C. REFERENCES

DOCUMENT	DOCUMENT TITLE
AMM	Aircraft Maintenance Manual
MTC	Standard Practices Manual

D. ABBREVIATIONS & DEFINITIONS

ABBREVIATION	DESCRIPTION
EC	Eurocopter (France)
ECL	Eurocopter Canada Limited
FWD	Forward
OTBD	outboard
PEI	polyetherimide
qty.	quantity
RH	Right-hand

E. UNITS OF MEASUREMENT

ABBREVIATION / SYMBOL	UNIT OF MEASUREMENT
hrs	hours
in	inch
kg	kilogram
lb	pound
m	meter

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2. AIRWORTHINESS LIMITATIONS

The Airworthiness Limitations section is approved by the Minister and specifies maintenance required by any applicable airworthiness or operating rule unless an alternative program has been approved by the Minister.

No airworthiness limitations associated with this installation.

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3. CONTROL AND OPERATION

Control and operation of the aircraft remains unchanged.

4. INSPECTION SCHEDULE AND MAINTENANCE ACTION

NOTE: Use torque per EC, MTC, Chapter 20.02.05.404 unless otherwise specified.

4.1. INSPECTION SCHEDULE

4.1.1. Every 100 flight hrs or 12 months (to coincide with the 100 hrs or 12 month helicopter inspection)

ITEM	INSPECTION OR MAINTENANCE WORK	CORRECTIVE ACTION
A	- Visually inspect door handle, in Figure 2 for: a. security	a. Secure as required.
B	- Visually inspect door handle in Figure 2 for: a. cracking	a. Cracking is not permitted. If cracking is evident, contact ECL for replacement part.
C	- Visually inspect RH side panel, shown in Figure 3 for: a. cracking around door handle attachment points	a. Cracking is not permitted. If cracking is evident, replace side panel in accordance with EC, Aircraft Maintenance Manual (AMM), Chapter 52-11-00, 4-1. Contact ECL for replacement part.

Table 1 Inspection Schedule and Maintenance Action
Every 100 flight hrs or 12 months

4.1.2. Every 600 flight hrs or 24 months (to coincide with the 600 hrs or 24 month helicopter inspection)

ITEM	INSPECTION OR MAINTENANCE WORK	CORRECTIVE ACTION
A	- Visually inspect outer reinforcement panel, inner reinforcement panel and spacer, shown in Figure 3. a. cracking b. corrosion	a. No cracking is allowed. If cracking is found, contact ECL for replacement part. b. No corrosion is allowed. If corrosion is found, contact ECL for replacement part.

Table 2 Inspection Schedule and Maintenance Action
Every 600 flight hrs or 24 months

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5. REPLACEMENT COMPONENTS AND REPAIR/OVERHAUL INFORMATION

No replacement components and repair/overhaul information required for this installation.

6. TROUBLESHOOTING

There are no unique characteristics which require troubleshooting techniques.

7. SPECIAL TOOLS

No special test equipment or tools are required. Standard tools are adequate.

8. REMOVAL AND REPLACEMENT

Proceed as follows if any of these items need to be removed:

PRELIMINARIES

- Read General Safety Instructions - Electrical Power Supply System (refer to EC 130 B4, Aircraft Maintenance Manual, Chapter 24-00-00, 3-1).
- Set the "D.BAT" pushbutton to "OFF" (refer to Removal / Installation - Battery, EC 130 B4, Aircraft Maintenance Manual, Chapter 24-33-00, 4-1).
- Set the "EXT PWR BAT" or "BAT EPU" (depending on MOD) pushbutton to "OFF" (refer to Electric Power Supply on the Ground EC 130 B4 Aircraft Maintenance Manual, Chapter 24-00-00, 2-1).
- Disconnect external power unit and the battery (refer to Removal / Installation - Battery, EC 130 B4, Aircraft Maintenance Manual, Chapter 24-33-00, 4-1).

A. REMOVAL

- 1) DOOR HANDLE (Refer to Figure 3)
 - a) Remove screws (3) (2 places) that secure the door handle to the inside of the RH Door.
 - b) Remove the door handle.

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8. REMOVAL AND REPLACEMENT (continued)

B. REPLACEMENT

NOTE: Use torque per EC, MTC, Chapter 20.02.05.404 unless otherwise specified.

- 1) DOOR HANDLE(Refer to Figure 3)
 - a) Reposition the door handle onto the inside of the RH door over the previously drilled holes.
 - b) Secure door handle using screws (3) (2 places).
- 2) Close all areas opened for service in the PRELIMINARIES paragraph of this section.
- 3) Before energizing the aircraft power supply system, read safety instructions (refer to Electrical Power Supply on the Ground, EC 130 B4, Aircraft Maintenance Manual, Chapter 24-00-00, 2-1).
- 4) Apply external power unit and battery. Refer to Removal/Installation EC 130 B4, Aircraft Maintenance Manual, Chapter 24-33-00, 4-1.
- 5) Perform functional tests in accordance with DC Power Supply System, Aircraft Maintenance Manual, Chapter 24-30-00,5-1.
- 6) Perform functional tests in accordance with EC 130 B4, Aircraft Maintenance Manual, Chapter 24-33-00, 5-1.

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9. WEIGHT AND BALANCE DATA

A. Removed Items

DESCRIPTION	WEIGHT		ARM		MOMENT	
	kg	lbs	m	in	kg m	lb in
(not applicable)	0.00	0.0	0.00	0.0	0.00	0.0
Total	0.00	0.0	0.00	0.0	0.00	0.0

B. Added Items

DESCRIPTION	WEIGHT		ARM		MOMENT	
	kg	lbs	m	in	kg m	lb in
Door Handle (qty. 1)	0.13	0.3	2.97	116.9	0.37	35.1
Spacer (qty. 2)	0.03	0.1	2.97	116.9	0.08	11.7
Outer Door Reinforcement	0.12	0.3	2.97	116.9	0.35	35.1
Inner Door Reinforcement	0.10	0.2	2.97	116.0	0.29	23.4
Total	0.37	0.9	2.97	116.9	1.09	105.2

10. PLACARDS AND MARKINGS

There are no placards or markings with this modification.

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